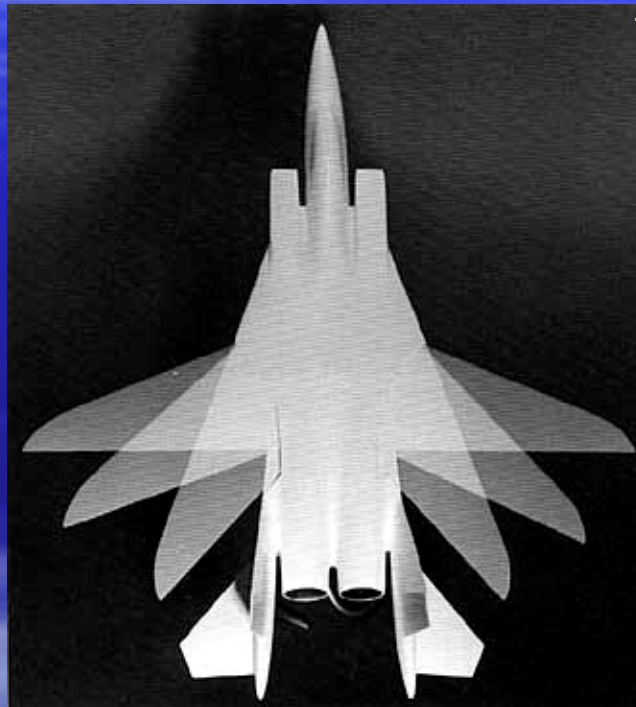


# Aerial Metamorphosis: Variable Sweep Wings



Brian Hayes  
Eugene Heim



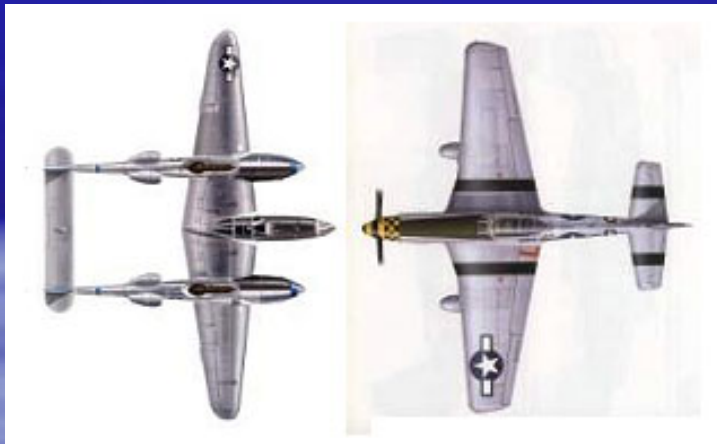
# Outline

- ▶ The Jet Age
  - ▶ Aerodynamic Lift At Supersonic Speed
  - ▶ Increasing Critical Mach Number
- ▶ In-Flight Optimization
  - ▶ Pivoting Around Stall
  - ▶ Variable Sweep – Variable Mission
- ▶ Variable Sweep Aircraft

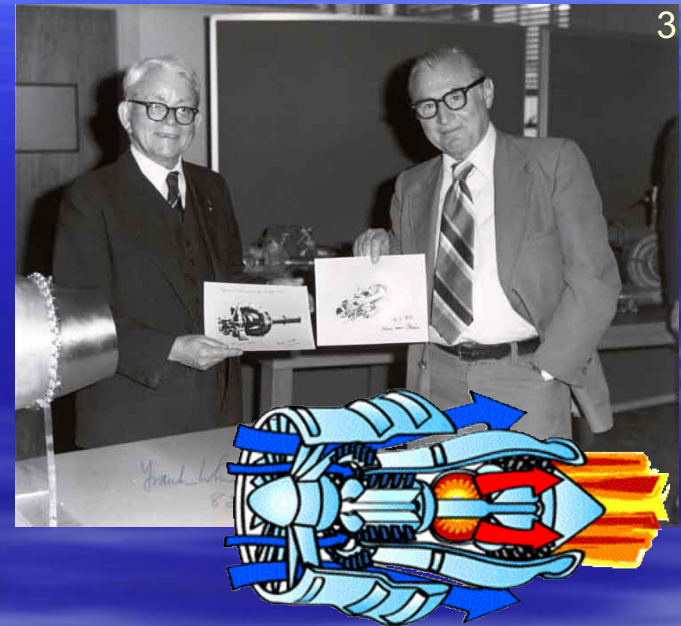


# The Jet Age

- Advances in propulsion in 1930's made sustained high-subsonic, transonic, and supersonic flight possible



[http://www.acepilots.com/planes/p51\\_mustang.html](http://www.acepilots.com/planes/p51_mustang.html)  
[http://www.acepilots.com/planes/p38\\_lightning.html](http://www.acepilots.com/planes/p38_lightning.html)

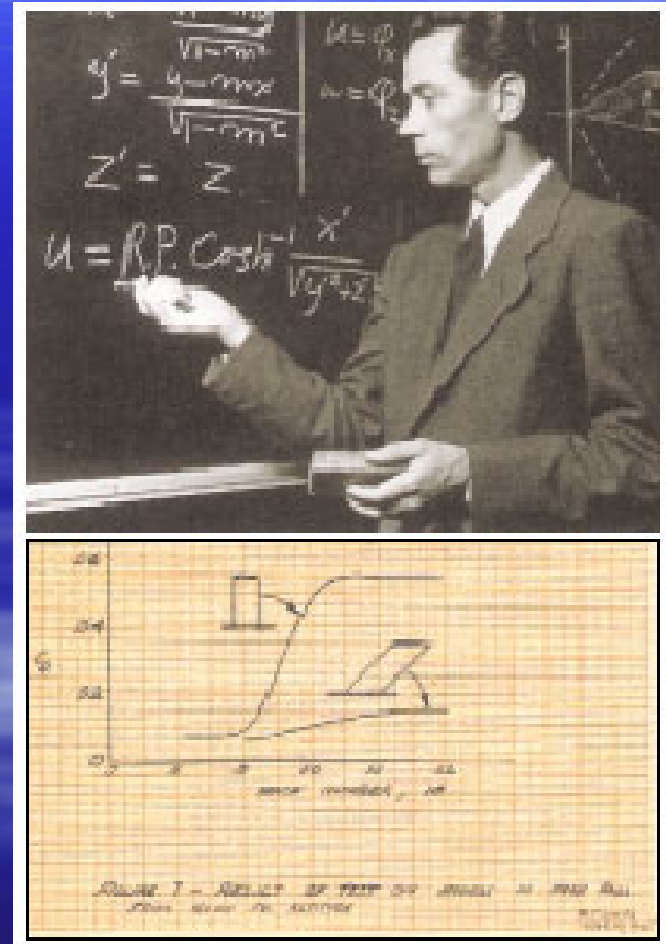


- High-speed WWII era aircraft typically had strait wings and thickness ratios of 14 to 18 percent



# Aerodynamic Lift At Supersonic Speed<sup>2</sup>

- 1935 Adolf Busemann, Aviation Research Organization, Germany
- 1941 Michael Gluhareff, Sikorsky chief of design
- 1945 Robert T. Jones, NACA Langley aerodynamicist

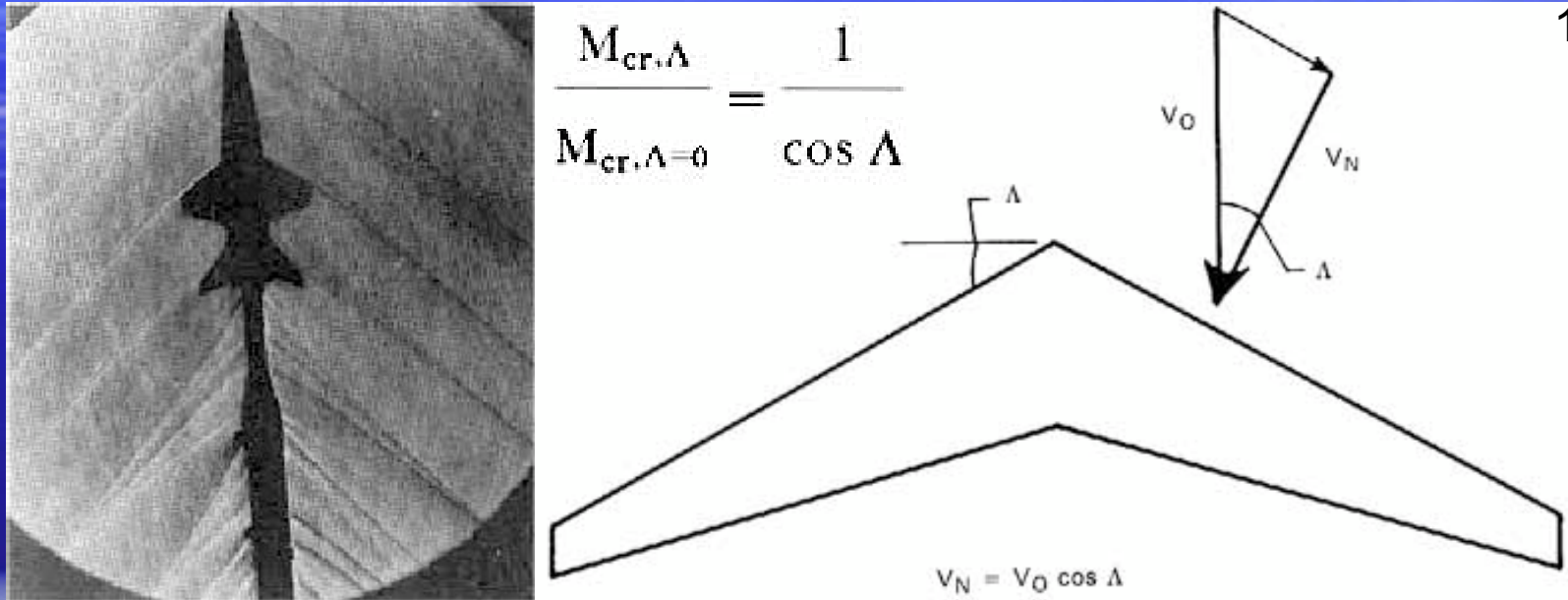


<http://www.hiller.org/about-us/briefings-journal/briefings/2000q4.pdf>



# Increasing Critical Mach Number

<http://www.eng.vt.edu/fluids/msc/gallery/shocks/p25b.htm>

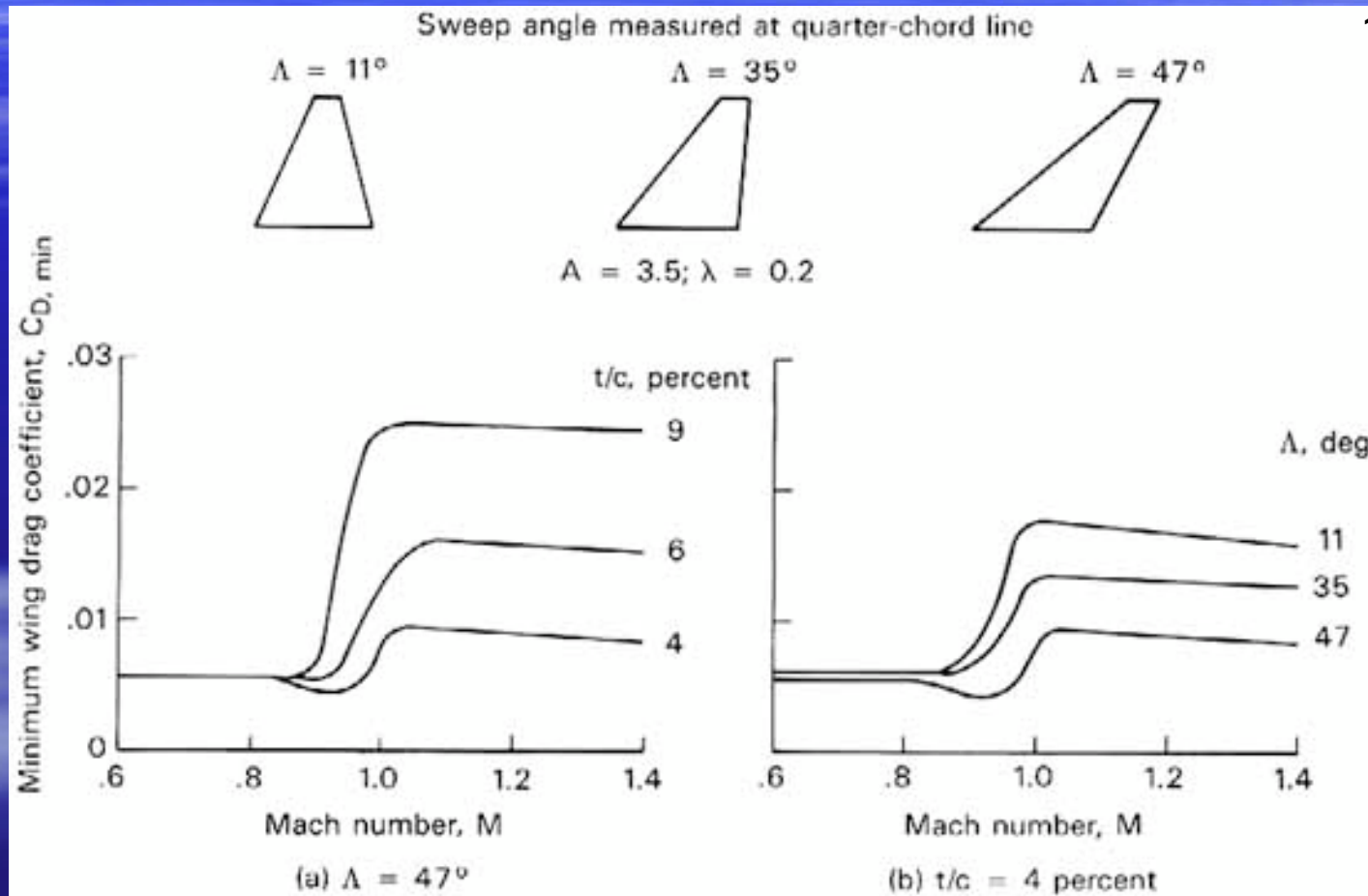


The composite image consists of three parts. On the left is a photograph of a dark wing against a light background, showing a series of diagonal lines representing shock waves. In the center is the equation 
$$\frac{M_{cr,\Delta}}{M_{cr,\Delta=0}} = \frac{1}{\cos \Delta}$$
. On the right is a diagram of a sweptback wing with a leading edge angle  $\Delta$ . A velocity vector  $V_0$  is shown approaching the wing, and its normal component  $V_N$  is shown perpendicular to the leading edge. The relationship  $V_N = V_0 \cos \Delta$  is written below the diagram. A small number '1' is in the top right corner of the diagram area.

- Independence of velocity components is true only for inviscid flow
- Overestimates  $M_{cr,\Delta}$



# Thickness Ratio and Sweep Angle

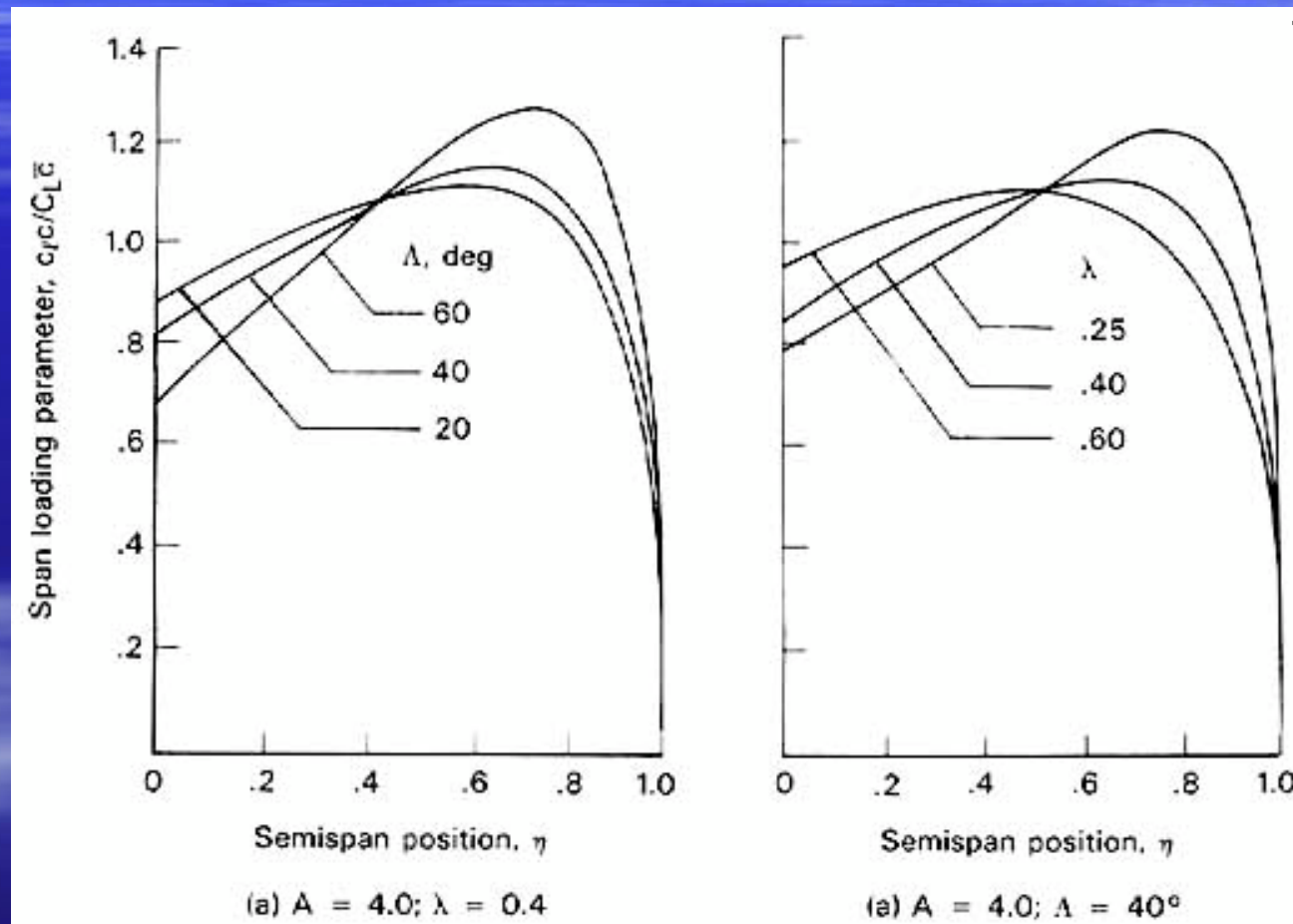


# Problems At Stall

- Serious S&C problems at high  $\alpha$  near stall
- NACA Bell L-39 swept wing flight study

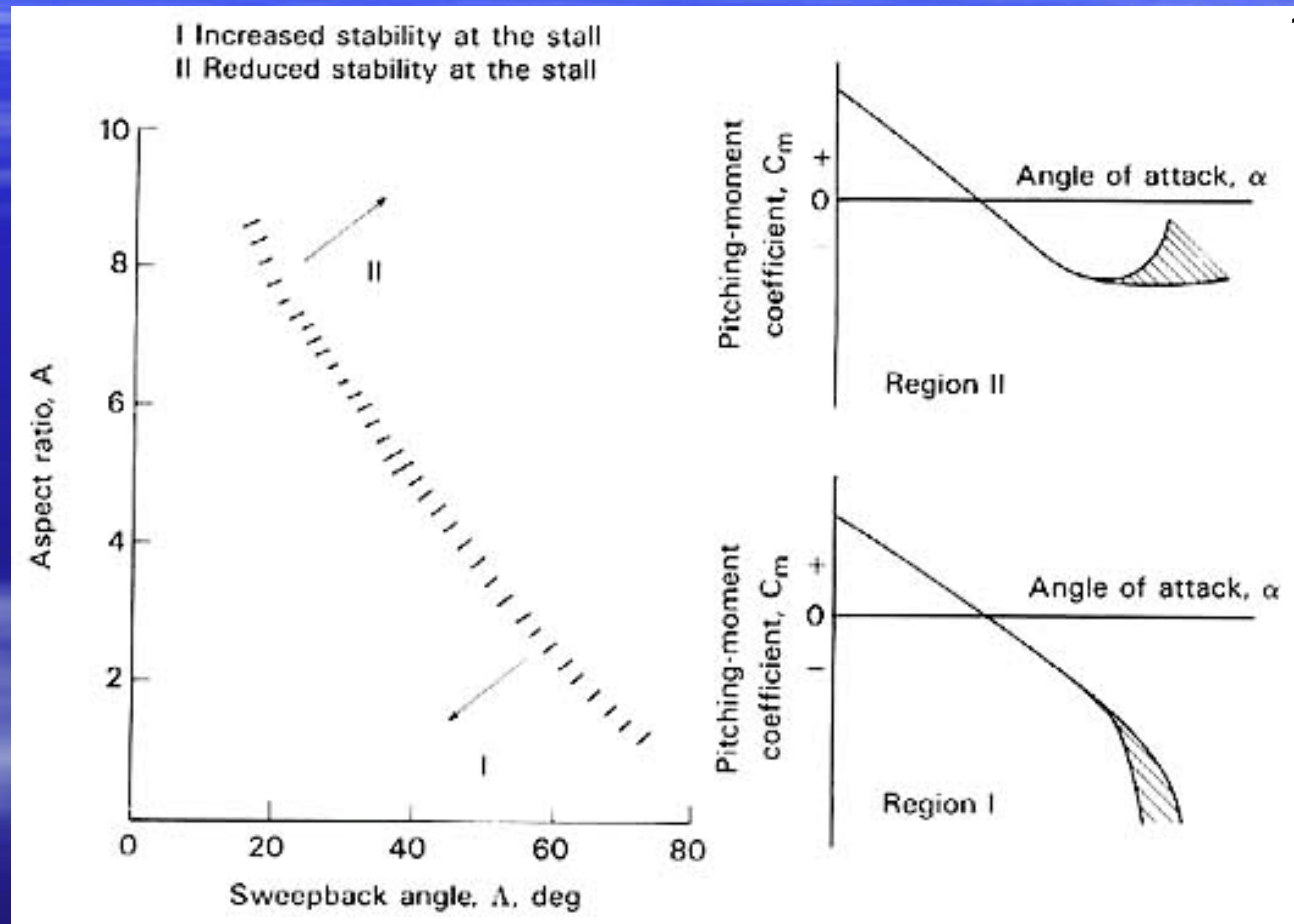


# Effect On Span Loading

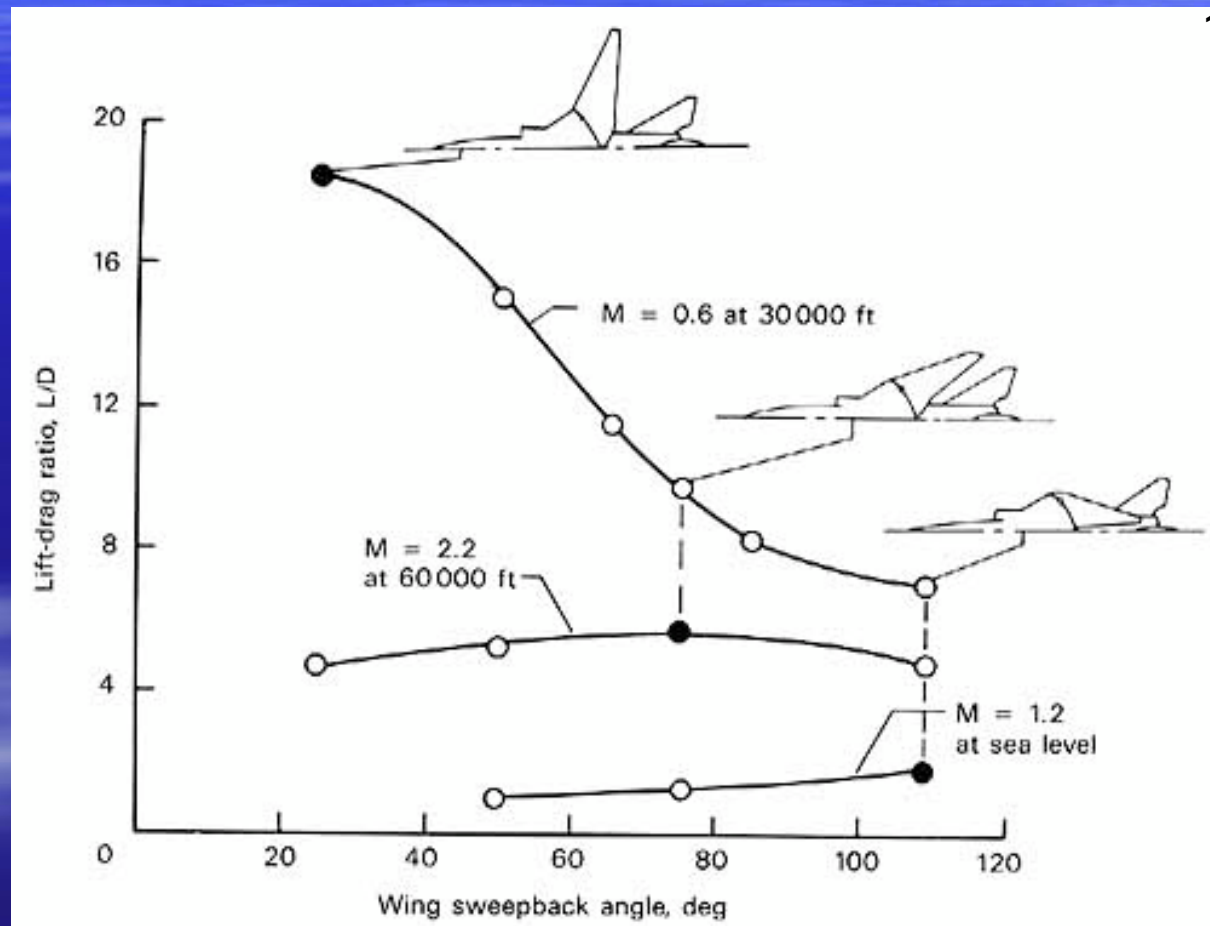




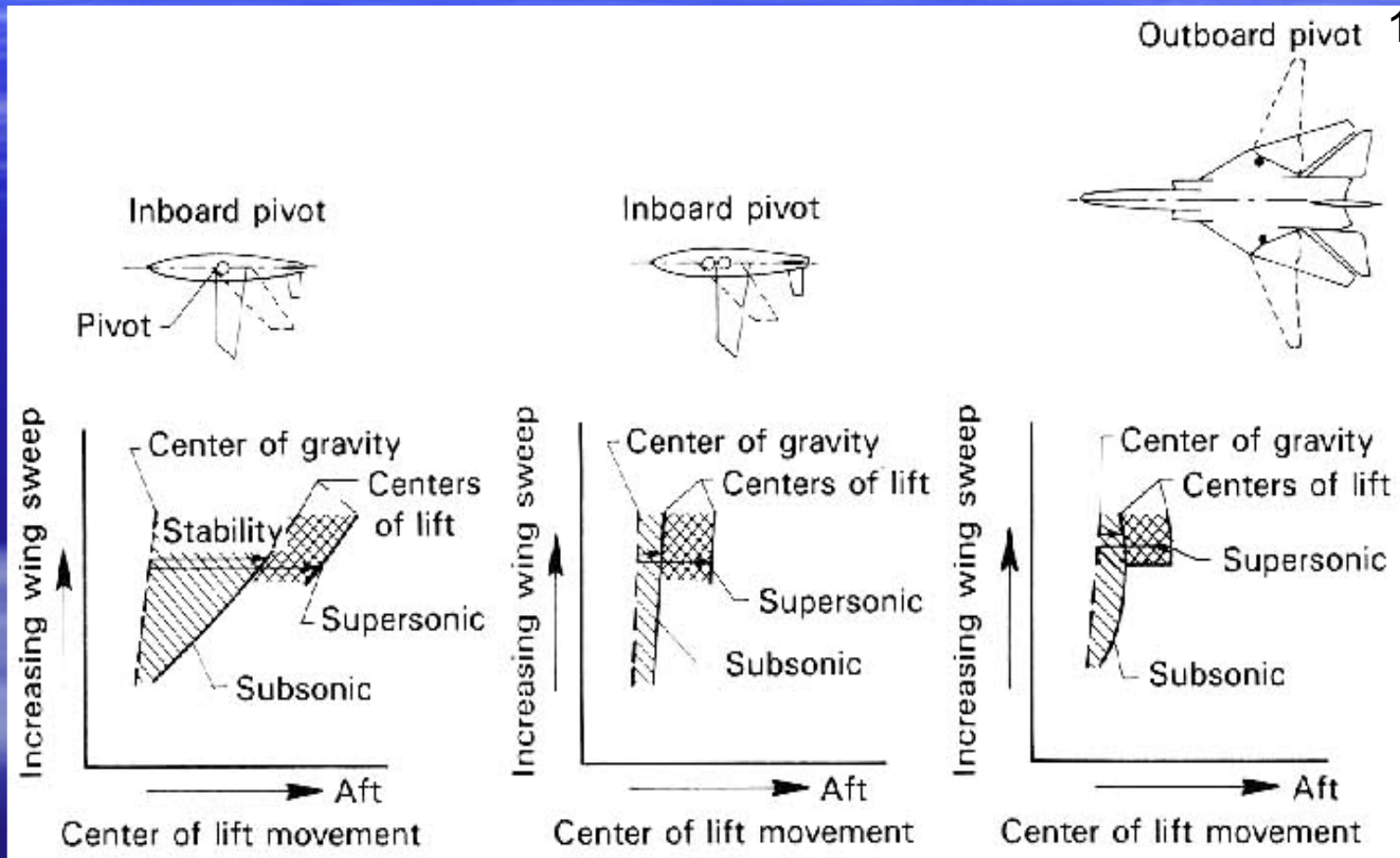
# Stall Stability Boundary



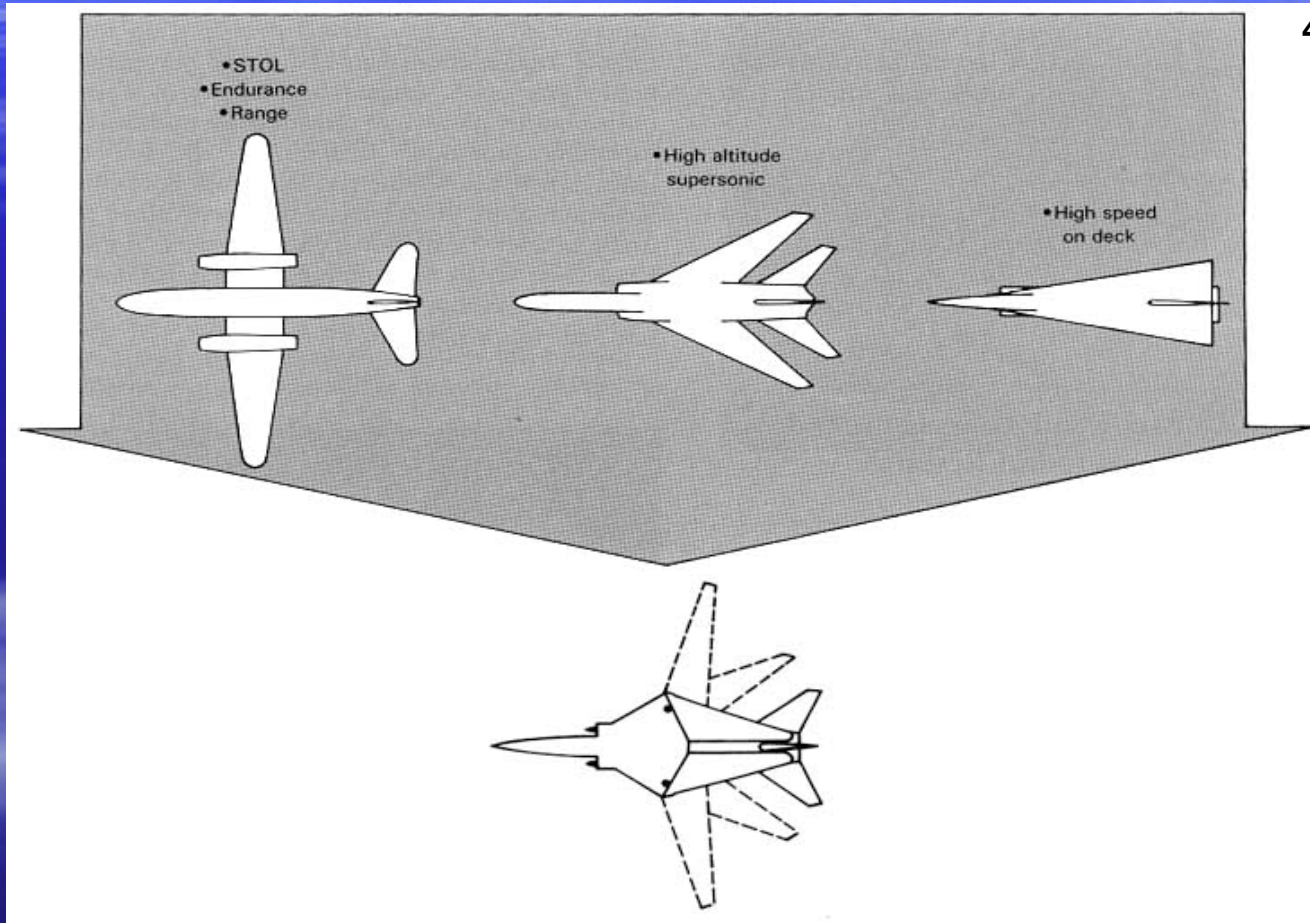
# In-Flight Optimization



# Pivoting Around Stability



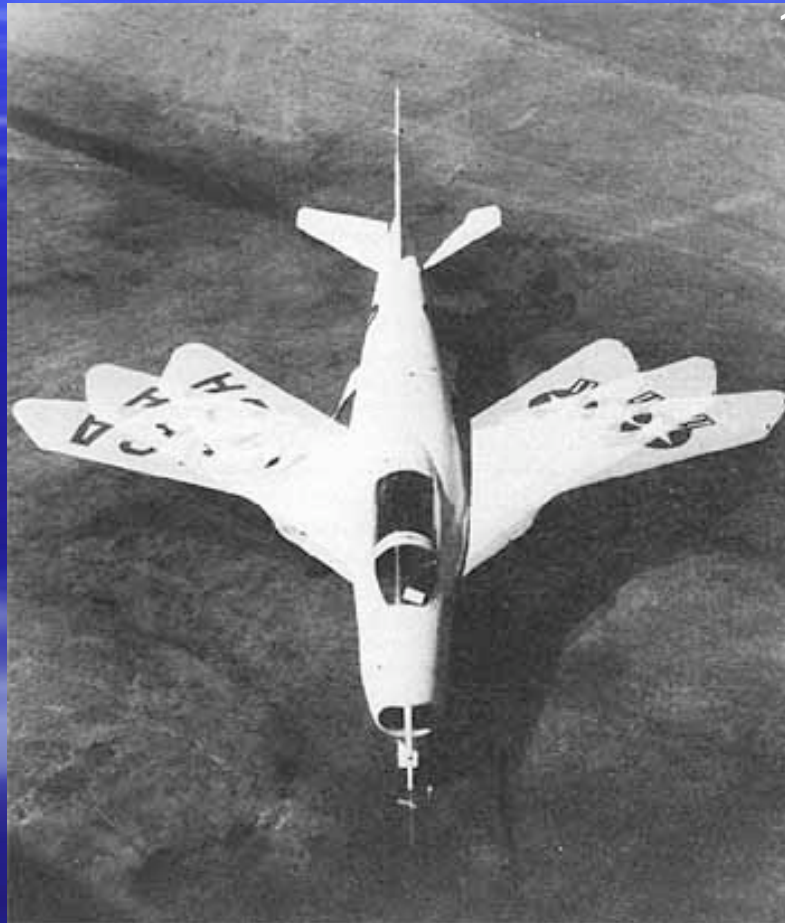
# Variable Sweep – Variable Mission



4



# Variable Sweep Aircraft



# Bell X-5

- ▶ Sweep varied from 20° to 60°
- ▶ Mach 0.81 showed dramatic performance differences for varied sweep
- ▶ Short field landing and take off
- ▶ High rate of climb
- ▶ Good loiter capability

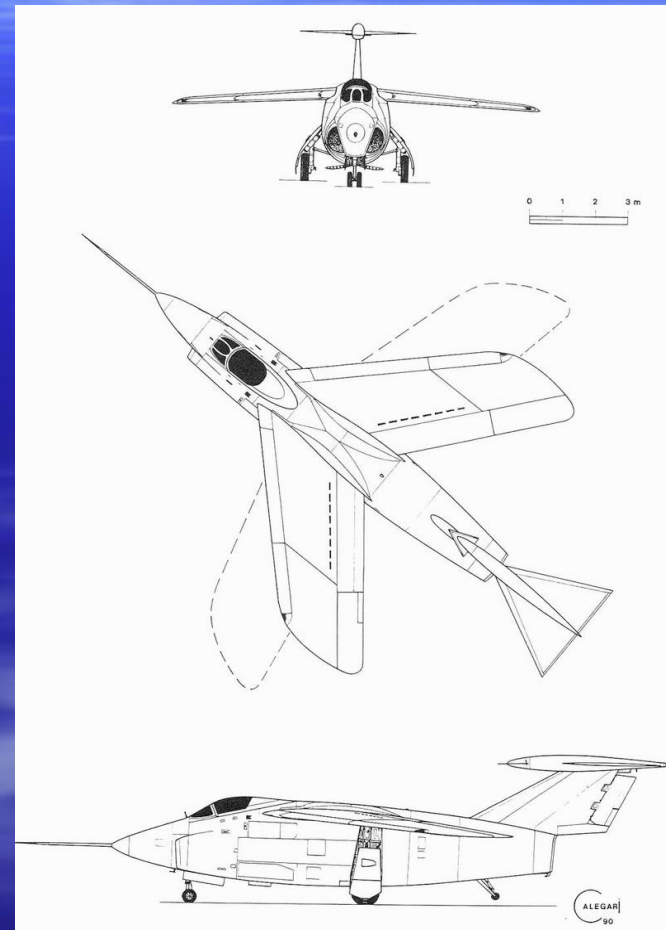


<http://users.dbscorp.net/jmustain/x5.htm>



# Grumman XF10F-1

- Sweep varied from  $13.5^\circ$  to  $42.5^\circ$
- Goals:
  - Reduce transonic drag
  - Low speed takeoff / landing for carrier use



<http://jpcollat.free.fr/xf10/xf10.php3>



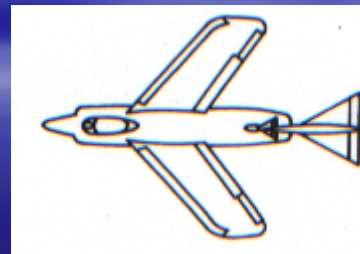
# XF10F-1 VS F3H-1

➤ Variable sweep provided:

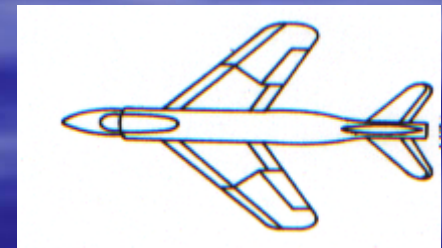
- Reduced landing speed
- Higher max speed
- Increased Range

|             | XF10F-1     | F3H-1  |                 |
|-------------|-------------|--------|-----------------|
| Weight      | 27,351      | 25,650 | lbs             |
| Wing area   | 467 - 450   | 442    | ft <sup>2</sup> |
| Span        | 36.8 - 50.6 | 35.3   | ft              |
| Stall speed | 78          | 96     | knots           |
| Max speed   | 617         | 592    | knots           |
| Range       | 1450        | 1025   | miles           |

XF10F-1



F3H-1



“Variable Sweep Wing Design”





# General Dynamics F-111

- Navy: carrier based
- Air force: long range
- Combination of requirements led to variable sweep design
- Sweep varied from  $16^\circ$  to  $72.5^\circ$



# Grumman F-14

- Carrier based air defense
- Low speed take off / landing
- Capable of Mach 2.4
- Sweep varied from 20° to 68°

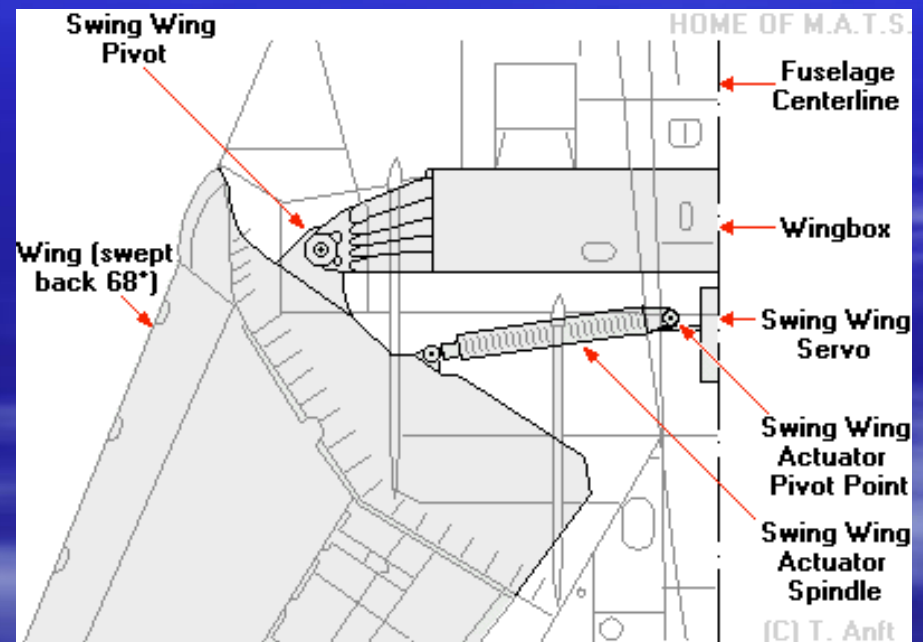


<http://www.anft.net/f-14/fl4-detail-wsm-04.htm>



# F-14 Sweep Mechanism

- Disadvantages:
  - Added complexity
  - Added weight
- Really add weight?
  - Larger wing area
  - Larger engines

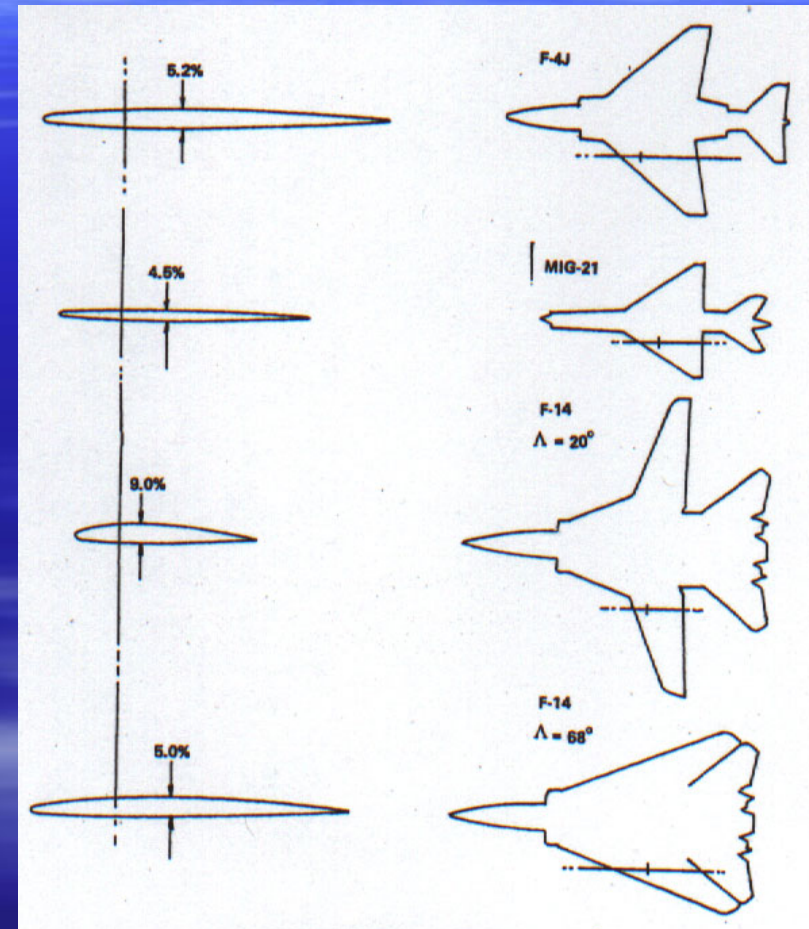


<http://www.anft.net/f-14/f14-detail-wsm-04.htm>



# F-14 Sweep Benefits

- Combines low and high speed performance
- Airfoil for subsonic and supersonic
- Reduced drag to weight ratio
- Lower thrust to weight required



“Variable Sweep Wing Design”



# Rockwell International B-1

➤ Sweep varied from  $15^\circ$  to  $67.5^\circ$

➤ Forward swept:

➤ Takeoff / landing

➤ High altitude cruise

➤ Aft swept:

➤ High subsonic flight

➤ Supersonic flight



<http://www.globalsecurity.org/wmd/systems/b-1.htm>



# Switchblade

- Possible design of forward variable sweep
- Leading edge becomes trailing edge
- Thought to be the now unveiled Boeing bird of pray



<http://www.area51zone.com/aircraft/switchblade.shtml>



# References

1. <http://www.hq.nasa.gov/office/pao/History/SP-468/cover.htm>
2. <http://www.hq.nasa.gov/office/pao/History/SP-4406/cover.html>
3. <http://inventors.about.com/library/inventors/bljetengine.htm>
4. <http://www.hq.nasa.gov/office/pao/History/SP-440/cover.htm>
5. <http://users.dbscorp.net/jmustain/x5.htm>
6. <http://www.vectorsite.net/avb1.html>
7. <http://www.area51zone.com/aircraft/switchblade.shtml>
8. <http://www.globalsecurity.org/military/systems/aircraft/f-111-design.htm>
9. <http://www.anft.net/f-14/index.htm>
10. <http://www.globalsecurity.org/military/systems/aircraft/f-14.htm>
11. “Variable Sweep Wing Design” by: Robert W. Kress
12. Bullets and lower right figure courtesy of <http://www.bestanimations.com/Military/Planes/F-14/F-14.html>
13. Virginia Tech logo courtesy of <http://filebox.vt.edu/vaes/HRAREC/images/>



# Questions?



<http://www.3dmodelworks.com/redir.asp?Page=thumb-show.asp%3FMODEL%3D706>





# Cool Picture



# Free Flight F-111

